

HIGH TIP SPEED FAN INLET NOISE REDUCTION USING TREATED INLET SPLITTERS AND ACCELERATING INLETS

(Quiet Engine Program Fan C Scale Model)

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S. B. Kazin

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TABLE OF CONTENTS

		Page
1.	SUMMARY	1
ii.	INTRODUCTION	4
	A. Background	4
	B. Vehicle Description	5
•	C. Test Program and Data Analysis	. 6
III.	INLET WITH MULTIPLE SPLITTERS	8
	A. Acoustic Data	8
•	1. Static 200-Foot (60.96 m) Sideline Results	. 8
	2. Flight Noise Results	10
	B. Aerodynamic Results	10-
	1. Mach Number and Recovery	10
	2. Aero-Acoustic Summary	11
.v.	INLETS WITH ONE SPLITTER	12
	A. Acoustic Data	12
	1. Static 200-Foot (60.96 m) Sideline Results	12
	2. Flight Noise Results	13
	B. Aerodynamic Results	13
	1. Mach Number and Recovery	13
	2. Aero-Acoustic Summary	14
v.	INLETS WITHOUT SPLITTERS	15
	A. Acoustic Data	15
-	1. Static 200-Foot (60.96 m) Sideline Results	15
	2. Flight Noise Results	16
	B. Aerodynamic Results	16
	1. Mach Number and Recovery	16
•	2. Aero-Acoustic Summary	16
VI.	CONCLUSIONS	17

TABLE OF CONTENTS (Concluded)

		rage
VII	NOMENCLATURE	18
	APPENDIX A - FIGURES	19
	APPENDIX B - ONE-THIRD OCTAVE DATA	88
	REFERENCES	121

LIST OF ILLUSTRATIONS

Figure		Page
1.	Performance Map, Baseline Bellmouth Inlet.	21
2.	Fan Vehicle Cross Section.	22
3.	Inlets Without Splitters.	23
4.	Inlets with Varying Splitters.	24
5.	Inlets with One Splitter.	25
6.	Fan Test Vehicle.	26
7.	Fan Test Facility.	. 27
8.	200-ft (60.96 m) Sideline PNL Vs. Angle from Inlet, Takeoff.	28
9.	200-ft (60.96 m) Sideline Δ SPL Vs. Frequency, 70°, Takeoff Referenced to Inlet with no Splitter.	29
10.	200-ft (60.96 m) Sideline SPL Vs. Frequency, 70° , Takeoff.	30
11.	200-ft (60.96 m) Sideline SPL Vs. Frequency, 120°, Takeoff.	31
12.	200-ft (60.96 m) Sideline PNL Vs. Angle from Inlet, 84% Fan Speed.	32
13,	200-ft (60.96 m) Sideline Δ SPL Vs. Frequency, 70° , 84% Fan Speed, Referenced to Inlet with no Splitter.	33
14.	200-ft (60.96 m) Sideline SPL Vs. Frequency, 70°, 84% Fan Speed.	34
15.	200-ft (60.96 m) Sideline PNL Vs. Angle from Inlet, 72% Fan Speed.	35
16.	200-ft (60.96 m) Sideline ASPL Vs. Frequency, 50°, 72% Fan Speed, Referenced to Inlet with no Splitter.	. 36
17.	200-ft (60.96 m) Sideline SPL Vs. Frequency, 50° , 72% Fan Speed.	37
18.	200-ft (60,96 m) Sideline PNL Vs. Angle from Inlet	38

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v

LIST OF ILLUSTRATIONS (Continued)

Figure	·	Page
19.	200-ft (60.96 m) Sideline ΔSPL Vs. Frequency, 50° Approach, Referenced to Inlet with no Splitter.	39
20.	200-ft (60.96 m) Sideline SPL Vs. Frequency, 50° , Approach.	40
21.	200-ft (60.96 m) Sideline SPL Vs. Frequency, 120° , Approach.	41
22.	Forward Maximum PNL.	42
23.	Aft Maximum PNL.	43
24.	1000-ft (304.8 m) Level Flyover PNL, Fan plus Jet Noise (Takeoff).	44
25.	1000-ft (304.8 m) Level Flyover PNLT, Fan plus Jet Noise (Takeoff).	45
26.	1000-ft (304.8 m) Level Flyover SPL, Fan plus Jet Noise (Takeoff, 70°).	46
27.	370-ft (112.8 m) Level Flyover, Fan plus Jet Noise (Approach).	47
28.	370-it (112.8 m) Level Flyover PNLT, Fan plus Jet Noise (Approach).	48
29.	370-ft (112.8 m) Level Flyover SPL, Fan plus Jet Noise (Approach, 60°).	49
30.	Average Inlet Throat Mach Number Vs. Corrected Fan Speed.	50
31.	Outer Wall Mach Distribution, Takeoff.	51
32.	Inlet Total Pressure Recovery Vs. Corrected Fan Speed.	52
33.	200-ft (60.96 m) Sideline Front Maximum PNL.	53
34.	200-ft (60.96 m) Sideline PNL Vs. Angle from Inlet, Takeoff.	54
35.	200-ft (60.96 m) Sideline SPL Vs. Frequency, Takeoff,	

The second of th

LIST OF ILLUSTRATIONS (Continued)

Figure		Page
36.	200-ft (60.96 m) Sideline PNL Vs. Angle from Inlet, 84% Fan Speed.	56
37.	200-ft (60.96 m) Sideline PNL Vs. Angle from Inlet, 72% Fan Speed.	57
38.	200-ft (60.96 m) Sideline PNL Vs. Angle from Inlet, Approach.	58
39.	200-ft (60.96 m) Sideline SPL Vs. Frequency, Approach, 70° .	59
40.	1000-ft (304.8 m) Level Flyover PNL, Fan plus Jet Noise (Takeoff).	60.
41.	1000-ft (304.8 m) Level Flyover PNLT, Fan plus Jet Noise (Takeoff).	61
42.	1000-ft (304.8 m) Level Flyover SPL, Fan plus Jet Noise (Takeoff, 70°).	62
43.	370-ft (112.8 m) Level Flyover PNL, Fan plus Jet Noise (Approach).	63
44.	370-ft (112.8 m) Level Flyover PNLT, Fan plus Jet Noise (Approach).	64
45.	370-ft (112.8 m) Level Flyover SPL, Fan plus Jet Noise (Approach, 60°).	65
46.	Average Throat Mach Number Vs. Corrected Fan Speed for Inlets with C. Splitter.	66
47.	Outer Wall Mach Distributi n, Takeoff.	67
48.	Inlet Total Pressure Recovery Vs. Corrected Fan Speed.	68
49.	200-ft (60.96 m) Sideline Front Maximum PNL, 90% Fan Speed.	69 .
50.	200-ft (60.96 m) Sideline Front Maximum PNL, Various Fan Speeds.	70
51.	200-ft (60.96 m) Sideline PNL Vs. Angle from Inlet, 88% Fan Speed.	71
52.	200-ft (60.96 m) Sideline SPL Vs. Frequency, 88%	5 0

LIST OF ILLUSTRATIONS (Concluded)

Figure		Page
53.	200-ft (60.96 m) Sideline PNL Vs. Angle from Inlet, 84% Fan Speed.	73
54.	200-ft (60.96 m) Sideline PNL Vs. Angle from Inlet, 72% Fan Speed.	74
55,	200-ft (60.96 m) Sideline PNL Vs. Angle from Inlet, Approach.	.· 75
56.	200-ft (60.96 m) Sideline SPL Vs. Frequency, Approach, 50°.	76
57.	200-ft (60.96 m) Sideline SPL Vs. Frequency, Approach, 120°.	77
58.	1000-ft (304.8 m) Level Flyover PNL, Fan plus Jet Noise, 88% Fan Speed.	78
59.	1000-ft (304.8 m) Level Flyover PNLT, Fan plus Jet Noise, 88% Fan Speed.	79
60.	1000-ft (304.8 m) Level Flyover SPL, Fan plus Jet Noise, 88% Fan Speed, 70°.	80
61.	370-ft (112.8 m) Level Flyover PNL, Fan plus Jet Noise, Approach.	81
62.	370-ft (112.8 m) Level Flyover PNLT, Fan plus Jet Noise, Approach.	82
63.	370-ft (112.8 m) Level Flyover SPL, Fan plus Jet Noise, Approach, 60°.	83
64.	Average Inlet Throat Mach Number Vs. Jorrected Fan Speed for Inlets Without Splitters.	84
65.	Outer Wall Mach Distribution, 85% Fan Speed.	85
66.	Inlet Total Pressure Recovery Vs. Corrected Fan Speed.	86
67	200 ft (60 96 m) Cidolino Pront Hardman DNI	97

I. SUMMARY

A series of inlet tests were run to determine the effects on noise of varying:

the number of inlet splitters

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- inlet acceleration with one splitter
- inlet acceleration with no splitters.

Each test was run with acoustic treatment which consisted of Scottfelt covered with a perforated plate. A total of eight suppressed configurations and an untreated baseline were run (see Figures 2-7).

The inlets were identified by their design takeoff average throat Mach number. Thus a 0.55 inlet implies a design intent of an average throat Mach number of 0.55 at 90% corrected fan speed. The fan was the scale model of the outer flowpath of the Quiet Engine Program's Fan C. This fan has a design point tip speed of 1550 ft/sec (472.44 m/sec) and a pressure ratio of 1.6.

A summary of the forward maximum PNL, throat Mach number, and total pressure recovery at takeoff and approach fan speeds is shown in Table 1. Some salient features of this table are:

- The 3 splitter inlet with Mach number acceleration from 0.46 (untreated baseline) to 0.67 reduced the noise 17.2 PNdB at takeoff with an inlet recovery loss of 2.9%. At approach, acceleration was from 0.26 to 0.35 with a noise reduction of 12.8 PNdB and a recovery loss of 0.7%.
- With one splitter the 0.79 inlet shows a reduction of 18.1 PNdB at takeoff with an acceleration of 0.46 to 0.72 and a recovery loss of 2.3%.
- When no splitters are employed, the reduction at takeoff in going from the untreated to the 0.55 inlet is 11.0 PNdB. With acceleration from 0.54 to 0.72 (see note f in the Table) a further reduction of 3.9 PNdB is realized. The total noise reduction, 14.9 PNdB, was obtained at a cost of 1.5% in recovery.

 Moderate levels of acceleration (i.e. from the 0.55 to the 0.65 inlet) increased the noise level.

In some instances more refined aerodynamic designs may reduce the losses measured with this test hardware.

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Table 1. Inlet Noise^a, Mach Number^b, and Recovery^c Summary

		Takeof f	d 	Approach							
Configuration	PNL	M _{TH}	η _r	PNL	M _{TH}	nr					
Unsuppressed	122.9	0.46	0.997	105.1	0.26	0.998					
0.55 no split.	111.9	0.54	0.994	98.5	0.30	0.998					
0.55 one split.	110.2	0.57	0.982	95.0	0.31	0.99					
0.55 two split.	108.6	0.61	0.982	94.7	0.33	0.99					
0.55 three split.	105.7	0.67	0.968	92.3	0.35	0.99					
0.65 no split.	112.9	0.62	0.992	99.4	0.33	0.99					
0.65 one split.	109.3	0.68	0.977	96.8	0.35	0.99					
0.79 no split ^f .	108.0	0.71	0.988	99.8	0.37	0.99					
0.79 one split.	104.8	0.72	0.974	96.7	0.37	0.99					
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- a. 200-foot sideline maximum forward angle full-scale PNL.
- b. Average throat Mach number based on flow and total pressure recovery.
- c. Average total pressure recovery.
- d. Takeoff is defined as 90% correc d fan speed.
- e. Approach is defined as 57.5% corrected fan speed.
- f. Data at 88% fan speed.

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II. INTRODUCTION

A. Background

The benefits of acoustically treating the inlet walls of turbofan engines has been clearly demonstrated. In fact, the inclusion of such treatment is an accepted part of any new commercial turbofan design. The continuing pressure for still lower noise levels, however, will require still greater efforts in inlet noise suppression.

There are at present two inlet noise suppression concepts being actively investigated. They are

- Multiple acoustically treated inlet splitters
- Choking the inlet flow

The latter is the newer of the two and as a result encompasses many as yet unexplored areas of operation. In general, the choked inlet will require some type of variable geometry.

The program whose results are examined in this report was designed to marry the two concepts in principle (i.e. treatment and high inlet Mach number-hybrid) while maintaining a fixed geometry inlet. In this manner the high Mach number exists at takeoff with the upper limit fixed by the requirement that the inlet be able to pass the altitude cruise flow. At lower power settings the Mach number is, of course, lower and the inlet treatment is then the sole means of suppression.

To cover the various design combinations, a test matrix was set up around three basic inlet cowl and lip combinations. Each such inlet was designated according to its ideal average throat Mach number at takeoff conditions. The Mach numbers chosen were 0.55, 0.65, and 0.79.

The 0.55 Mach number is representative of conventional design practices and forms the baseline for the test series. A 0.65 Mach number represents about the upper limit for a conventional takeoff and landing aircraft since such an inlet will just pass the altitude cruise flow. Finally, the 0.79 inlet could be used for a STOL aircraft where the engine is sized at takeoff rather than at the altitude cruise or with some variable geometry feature.

A single inlet splitter was also designed for each of these cowls. And to fully examine the splitter concept, two and three splitter arrangements were designed for the 0.55 cowl and inlet lip.

Thus three basic geometric arrangements over a range of Mach numbers were investigated.

- 0.55, 0.65, and 0.79 average throat Mach numbers in a wall treated inlet
- The above three cowls with single treated splitters
- The 0.55 cowl with 0, 1, 2, and 3 treated splitters.
- B. Vehicle Description

15 Carry 7.15

As a basic test vehicle the scale model Fan C of the Quiet Engine Program was utilized. Some of the basic characteristics of this fan are listed below.

Design P/P - 1.6

Design tip speed - 1550 ft/sec (472.44 m/sec)

Number of blades - 26

Number of vanes - 60

Rotor - OGV spacing - 2 rotor tip chords

Takeoff % corrected speed - 90

Approach % corrected speed - 57.5

Tip diameter - 36 inches (91.44 cm)

Radius ratio - 0.57

The high radius ratio results since the scale model fan represents only the bypass flowpath portion of a turbofan engine with a design bypass ratio of 5.0. Figure 1 is the performance map of the fan as determined with a clean inlet bellmouth (no stall line testing was done on this vehicle).

For these tests a blade shape designated "Mod II Blade" was employed. A series of tests with different blade shapes will be the subject of a later report.

A cross section of the basic vehicle is shown in Figure 2. This sketch shows the fan with the 0.55 cowl and bell lip installed. The acoustic treat-

ment consists of one-half nch 2-900 Scottfelt covered by a perforated place with an open area of 22-1/2%. The holes are 1/16 inch (1.53 mm) in diameter and the faceplate is 0.03 inches (0.76 mm) thick.

Since this test series was to be an evaluation of front end noise, the long suppressor was added to the rear to remove rear radiated fan noise from the front farfield quadrant.

Figures 3-5 show the various inlets. In each case the cowl treatment extends 29 inches upstream of the leading edge of the fan rotor. In full scale (22,000 pound thrust engine) this would represent 53 inches (134.62 cm) of treatment. The acoustically treated splitter sets for the 0.55 inlet were optimized for each of the configurations. That is, the one splitter case was set where it would intercept the most acoustic energy and yet not be too far away or close to the outer wall for good suppression characteristics. The same holds true for the two and three splitter arrangements. In other words the decreasing numbers of splitters were not obtained by removing splitters from the 3 splitter inlet, but rather by designing new splitters and support struts.

The cowl for the 0.65 and 0.79 inlets was the same. The higher Mach number was obtained by a change in the bell lip. This is the reason that the 0.79 inlet is slightly longer than the other two inlets. As a result of this, the same single splitter was used for the 0.65 and 0.79 inlets.

It should be noted that the splitters actually extend into the throat so that the inclusion of splitters also raises the average throat Mach number. This resulted from application of the design criteria that the inlet should contain as much acoustic treatment as possible. It is possible, however, as the results will show that this added acoustic benefit was paid for in higher inlet losses.

C. Test Program and Data Analysis

The test vehicle is shown set up on the test stand in Figure 6. The facility is driven by an LM1500 gas generator connected to the fan vehicle through an inlet shaft. The inlet bell lip is about 28 inches (71.1 cm) away

from the first shaft bearing pedestal. It is possible that this arrangement has an effect on the absolute level of noise; however, since each configuration was run under the same conditions the relative evaluation of the inlets is valid.

Figure 7 is a view of the test facility. The sound field consists of a 100-foot arc of microphones spaced at 10 degree increments from 20 to 160 degrees. The microphones were placed at the fan centerline height - 12.5 ft. (3.81 m) off the ground. The field itself is covered with asphalt.

Data is FM recorded on magnetic tape at 60 inches per second (152.4 cm/sec) in the control room. These data are then analyzed on a General Radio 1/3 - octave analyzer using a 32 second averaging time. The analyzed data is corrected to standard day. At this point the data is still in scale model size. However, a more realistic evaluation can be made if the data is scaled to full size. This is done by shifting the spectrum down in the same ratio as the ratio of the scale model's blade passing frequency to the full scale's blade passing frequency. In addition, an adjustment for the size is made by adding 10 log of the weight flow ratio to all of the data. The linear scale factor is .527.

Unless otherwise noted all data presented in this report is scaled to full scale and projected to the 200-foot (60.96 m) sideline.

The typical test program sequence consisted of running along the fan's nominal operating line taking farfield noise data at various intervals for one and one-half to two minutes at each speed point. This process was then repeated so that all data represents the average of a run and repeat.

Aerodynamic testing was done on separate runs so that all aerodynamic instrumentation could be removed during acoustic testing. Aerodynamic instrumentation consisted of boundary layer rakes, static pressures, and total and static pressure traverses. From these data and calculations the recovery and Mach number distributions were determined.

III. INLET WITH MULTIPLE SPLITTERS

A. Acoustic Data

1. Static 200-Foot (60.96 m) Sideline Results

In order to assess the effects of the addition of splitters to the inlet along with increasing the throat average Mach number, splitters were added to the 0.55 design Mach number inlet. The splitters were added one at a time up to 3 splitters. Cross-sections of these inlets are shown in Figure 4. In each case the same cowl and bellmouth were used but the splitters were placed so as to provide the maximum suppression for the number of splitters being employed.

Figure 8 shows the resulting reduction at takeoff (90% corrected fan speed) for the treated cowl and successive addition of splitters. One of the more noteworthy aspects of these data is the small difference between one and two splitters - about 1-1/2 PNdB at 70 degrees - compared to the overall reduction of 6 PNdB. Figure 9 contains a comparison of the 1/3-octave spectra at 70 degrees referenced to the inlet with no splitters. The blade passing frequency is at 2000 Hz. Clearly the reduction obtained from 2 to 3.15 KHz accounts for the success of the three splitters. The one and two splitter data show very little difference all across the spectrum. The hump in suppression at 316 Hz is multiple pure tones (MPT) suppression. These data indicate that the inclusion of the first splitter gets the MPT suppression and that there is essentially no change with the addition of more splitters.

As a point of reference Figure 10 shows the 0 and 3 splitter absolute levels. The total suppression is maximum at the 2 to 2.5 KHz frequencies - about 10 dB. Also included is the 120 degree spectra for 0 and 3 splitters in Figure 11. The flat characteristic of this spectrum and the small level of reduction indicates that the rear angles are probably dominated by aft radiated noise caused by the jet and the air flowing over the internal aft splitter.

Figures 12-17 contain similar data for 84 and 72% corrected fan speed. At 84% speed the results have the same characteristics as at takeoff. At 72% speed, Figure 15 shows a peak at 50 degrees and a corresponding large

noise reduction as splitters are added. This particular speed is the speed at which the rotor tip relative Mach number exceeds one. A characteristic of the onset of supersonic relative fan tip speed is an increase in the blade passing frequency (BPF) level as shown in Figure 17. (At slightly higher fan speeds multiple pure tone noise develops.) Figure 16 indicates that the relatively large noise reduction obtained at 50 degrees is due to suppression of the high BPF level.

The data at approach fan speed is contained in Figures 18-21. Figure 18 indicates that the largest improvement was obtained with the addition of the single splitter; although successive improvements with additional splitters were obtained at 50 degrees. The spectral details contained in Figures 19 and 20 at 50 degrees shows that the most suppression has been obtained at the BPF in the 1250 Hz band.

It can be seen in Figure 18 that the suppression has extended around to the rear more than in the higher speed cases. This is probably due to the lower jet and internal scrubbing noise at 57.5% speed. The spectra for the 0 and 3 splitter cases are shown in Figure 21. Clearly the suppression at the BPF and its second harmonic account for the observed PNL decrease at 120 degrees.

Figures 22 and 23 show the progression of the 200-foot (60.96 m) sideline PNL with fan speed. The forward maximum PNL, Figure 22, shows the "jump" in level with no splitters at 70 to 72% speed is removed when the splitters are employed. In the rear quadrant, Figure 23 shows a family of data which progresses almost linearly with speed in all cases. The inlet data, however, shows a considerable flattening at high speeds particularly with the 3 splitters.

As was previously mentioned one of the design criteria was to place as much treatment in the inlet as possible. This resulted in a throat area decrease and corresponding throat Mach number increase in each case. At higher power settings therefore noise reduction was not only a result of treatment but also inlet Mach number increases. This is, in some measure, the reason for the leveling off at high speeds seen in Figure 22. That is,

as the fan speed increases so does the inlet Mach number. At speeds above 85%, particularly with 3 splitters, the increasing source level and increasing Mach number tend to offset each other resulting in no appreciable noise increase.

2. Flight Noise Results

In order to obtain a view of the noise reduction capabilities of these configurations in flight, the scaled data was "flown" through level flyovers at approach (57.5% fan speed) and takeoff (90% fan speed) power. The take-off flight altitude was at 1000 feet (304.8 m) and the approach at 370 feet (112.8 m). A flight Mach number of 0.22, temperature of 77° F, and a 70% relative humidity were employed.

A core jet was also added to the noise spectrum by using the method of SAE AIR 876. The SAE relative velocity correction was also applied to the low frequency noise. For the purpose of these comparisons this method is deemed acceptable. However, it is recognized that a greater degree of sophistication may be required to more accurately describe the absolute level of jet noise.

Figures 24 and 25 show, respectively, the takeoff PNL and PNLT for each configuration. The tone corrected data show approximately the same reduction as the uncorrected levels for 3 splitters. Figure 26 contains the spectra at 70 degrees for 0 and 3 splitters. The linear region below 315 Hz is the result of linearly extending the fan noise over the region of the spectrum where the relative velocity correction has reduced the jet noise.

The PNL and PNLT data at approach are shown in Figures 27 - 29. In this case the tone corrected data indicates a larger reduction. The spectra in Figure 29 clearly show the large BPF tone with no splitters which results in a tone correction that adds 4.7 PNdB at 40 degrees.

B. Aerodynamic Results

1. Mach Number and Recovery

As was previously mentioned the splitters were added such that the inlet throat was smaller in each case. Figure 30 shows the average throat Mach number trend with corrected fan speed. This Mach number was computed from the total measured flow and the measured inlet recovery. At higher

speeds the Mach number is considerably higher than the conventional inlet (usually .5 to .55). As will be shown in Section VI Mach numbers of about .65 and higher result in noise reduction solely because of the acceleration.

The average throat Mach number is a convenient correlating parameter; however, also of interest is the outer cowl surface Mach number. Figure 31 shows the trend in this parameter for each inlet. In the region of the throat all the configurations show a peak; although it tends to be lower as the splitters are introduced. This is due primarily to the loss of recovery at the takeoff corrected fan speed. In the case of the 3 splitter inlet, however, a "second throat" has appeared in the region of the splitter support struts.

The inclusion of splitters resulted in noise reduction at the cost of inlet recovery. Figure 32 shows the recovery versus corrected speed. With no splitters the inlet behaves in the normal manner with recovery at .994 at 90% speed. As splitters are added recovery drops with a low being measured with 3 splitters at .962 (Mach number is about .7). The one splitter inlet shows lower recovery than the 2 splitter inlet except at I'gh speed. It is believed that this loss in recovery was the result of a misalignment of the single splitter with respect to the flow. The effect of this on the noise reduction obtained is unknown; although it does not appear to have caused any discontinuities in the acoustic data.

2. Aero-Acoustic Summary

Finally, Figure 33 shows the recovery and noise trend at takeoff fan speed. As would be expected the noise decrease is paid for in inlet recovery loss. Roughly, a 1% decrease in recovery results in a 2% thrust loss on the Engine C cycle. Therefore, recovery levels as shown in Figure 33 will have to be carefully considered in the engine suppression design.

IV. INLETS WITH ONE SPLITTER

A. Acoustic Data

1. Static 200-Foot (60.96 m) Sideline Results

In this section data from a series of tests on the 3 cowl designs is examined with one acoustically treated splitter in the inlet (see Figure 5).

Figure 34 shows the takeoff PNL directivity for the 3 inlets. The 0.55 and 0.65 cowls produce maximum front levels which are about one PNdB apart; however, the 0.79 cowl shows a marked drop in level. This drop is due to the higher Mach number in the 0.79 inlet (0.72 versus 0.57 and 0.68 for, respectively, the 0.55 and 0.65 cowls) since all three inlets contain the same amount of treatment. At 70 degrees the difference between the 0.79 inlet and the other two configurations is 4 PNdB. The spectra, Figure 35, shows a hierarchy of noise level at frequencies between 2 KHz and 6350 Hz which is indicative of the Mach number effect.

As speed is decreased from takeoff Figures 36 through 38 show a marked change in this picture. When the Mach number decreases from the level at takeoff the 0.55 cowl produces the lowest noise and the 0.79 inlet produces the most noise. This is particularly true at approach (Figure 38, 57.5% speed). At the front maximum, 70 degrees, the 0.79 inlet is about one PNdB higher than the 0.65 inlet which is in turn about 2 PNdB higher than the 0.55 inlet. For the approach case the average throat Mach numbers are:

Inlet	TH av.
0.55	0.31
0.65	0.35
0.79	0.37

Although the Mach number range is small, it is possible that higher Mach numbers are generating higher inlet turbulence levels which create more noise without an attendent acceleration effect.

Figure 39 shows the spectral characteristics at 70 degrees. The noise increase spreads all across the spectrum including the blade passing frequency (1250 Hz).

In summary the 200-foot (60.96 m) sideline results show a marked acceleration effect at takeoff. However at approach the higher Mach number has acted to increase the noise level.

2. Flight Noise Results

The scaled static results were extrapolated to flight conditions and a core jet added to the spectrum (see Section IV.A.2). Figures 40 and 41 show the takeoff PNL and PNLT at 1000 feet (304.8 m). The tone corrected data show a still greater effect or Mach number than PNL. Spectral comparisons, Figure 42, show that the 0.79 inlet has no significant tone content as well as a lower level.

At approach power at 370 feet (112.8 m) the PNL and PNLT directivities are shown in Figures 43 and 44. To some extent the altitude accentuates the higher Mach number inlet problems. The spectra at 60 degrees, Figure 45, shows a very high tone content at this angle (other angles indicate a somewhat smaller increase).

B. Aerodynamic Results

1. Mach Number and Recovery

As has been noted the Mach number has played an important part in the noise reduction obtained with the 0.79 inlet. Figure 46 contains the average throat Mach number variation with corrected fan speed. At 95% speed the 0.79 inlet line "falls over". This is indicative of choking in at least part of the inlet. Figure 47 clearly shows that the wall Mach number is supersonic in the region of the throat with a rapid deceleration in the passage containing the splitter.

Figure 48 shows the inlet total pressure recovery versus corrected fan speed. At high speeds the 0.79 inlet is dropping rapidly. This is probably due to the choked outer flow passage and attendent shock and boundary layer losses.

2. Aero-Acoustic Summary

It is obvious that the Mach number effect which reduces the noise has come at the cost of inlet recovery. Figure 49 shows the noise trend with inlet recovery at 90% speed. The noise decrease due to the higher Mach number is about 5-1/2 PNdB but it was obtained at the cost of about 0.7% in inlet recovery. The absolute level of recovery for even the 0.55 inlet is also somewhat low - about 98.1%. It is, however, possible that some clean-up could be achieved by further iterations on the aerodynamic design. This is particularly true for the 0.79 inlet where some reduction in Mach number might be in order.

Figure 50 shows a "map" of the Mach number and inlet noise at various speeds. The tendency for the noise to increase with Mach number at lower speeds can be seen. At 88% speed the 0.55 and 0.65 inlet produce about the same noise level. Above this speed the 0.65 inlet is lower in noise. At 95% _pc.2d the 0.65 and 0.79 inlets produce about the same noise level at about the same Mach number.

V. INLETS WITHOUT SPLITTERS

A. Acoustic Data

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1. Static 200-Foot (60.96 m) Sideline Results

This test series provides data on the effects of higher inlet Mach numbers without acoustic splitters but with a treated nacelle wall. The cowl wall hardware was the same as in previous tests with splitters.

Figure 51 shows the PNL at 88% fan speed. The normal takeoff speed, 90%, was not used since the 0.79 inlet showed signs of flow separation at this speed. However these data are representative of high speed fan operation. The 0.79 inlet is clearly the lowest noise in Figure 51. However, the problem with the 0.65 inlet cited in the previous section is again apparent. The average throat Mach number for each inlet at this speed is:

Inlet	TH av.
0.55	0.52
0.65	0.60
0.79	0.71

At 70 degrees the 0.79 inlet shows a reduction of 2-1/2 PNdB over the 0.55 inlet; but the 0.65 inlet exceeds the 0.55 inlet by 2-1/2 PNdB at this angle. The spectra at 70 degrees are shown in Figure 52. With the 0.79 inlet the noise has decreased from the BPF at 2000 Hz to 6300 Hz while the 0.65 inlet noise shows increases over most of the spectrum relative to the 0.55 inlet.

As the fan speed is decreased to the approach speed (57.5%) the noise of the 0.79 inlet increases relative to the other two inlets until at approach the 0.79 inlet exceeds the other inlets at some angles (Figures 53-55). The spectra at 50 degrees (Figure 56) shows the hierarchy at the BPF (1250 Hz) to be 0.79, 0.55, and 0.65; however the 0.55 inlet produced the lowest noise at higher frequencies. The spectra at 120 degrees, Figure 57, shows about the same characteristics as the 50 degree spectra.

This increase in noise up to about 0.6 Mach number was also seen with the 0.65 inlet with a single splitter. Again it appears that moderately high Mach numbers increase noise rather than decreasing it.

2. Flight Noise Results

The flight noise data for 88% speed is contained in Figures 58 - 60. Generally the flight noise follows the conclusions drawn from the sideline; although the 0.55 and 0.65 inlets are nearly equal (in Figure 51 the 0.65 inlet was higher).

For approach power, Figures 61 - 63, the 20'-foot (60.96 m) sideline conclusions also hold. Figure 63 shows the 60 degree spectra with the direct Mach number-noise hierarchy at frequencies above the BPF.

B. Aerodynamic Results

1. Mach Number and Recovery

Figure 64 contains the corrected fan speed versus average throat Mach number characteristics for the 3 inlets. The outer wall Mach number distributions are shown in Figure 65. Generally these results are as would be expected.

In this case the recovery versus fan speed from inlet-to-inlet (Figure 66) follows the trend of higher Mach number lower recovery. The data for the 0.79 inlet do not go beyond 88% speed due to data acquisition problems at higher speeds. Extrapolation of this line may be misleading as it is suspected that the recovery is dropping at an accelerating rate.

2. Aero-Acoustic Summary

Figure 67 shows the noise and average throat Mach number characteristics of each inlet. These curves show the trend to higher noise with Mach number at lower fan speeds. Above 72%, however, the 0.79 inlet begins to produce lower noise.

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VI. CONCLUSIONS

- Multiple acoustically treated splitters and high average throat Mach number (0.67) result in appreciable takeoff noise reduction [17.2 PNdB on the 200-foot (60.96 m) sideline]; however inlet recovery at takeoff fan speed is down to 0.968.
- 2. The use of high inlet Mach number (0.72) with one acoustically treated splitter shows a takeoff noise reduction of 18.1 PNdB with a recovery of 0.974. Thus reduction in the number of splitters slightly improves both noise and inlet aerodynamic performance when higher Mach number is employed.
- 3. With an acoustically treated cowl, high inlet Mach number (0.71) shows a high fan speed noise reduction of 14.9 PNdB. Examination of these data indicate that at least 3.7 PNdB of this reduction is due to acceleration. Inlet recovery was 0.982. Thus the inlet without splitters showed the best ratio of recovery loss to PNL reduction 0.1% loss in recovery per PNdB.
- 4. Moderate increases in average throat Mach number (≤0.6) with treated inlets results in a noise increase. Noticeable acceleration effects appear at Mach numbers ≤ 0.65.

VII. NOMENCLATURE

BPF Blade passing frequency

Hz Hertz

M Mach number

Mth Average throat Mach number

M Flight Mach number

MPT Multiple pure tone (shock noise)

OGV Outlet guide vane

P/P Pressure ratio

PNdB Perceived noise decibel
PNL Perceived noise level

PNLT Tone corrected perceived noise level

RE "referenced to"

SL Sideline

SPL Sound pressure level

W/O "without"

n Total pressure recovery

APPENDIX A - FIGURES

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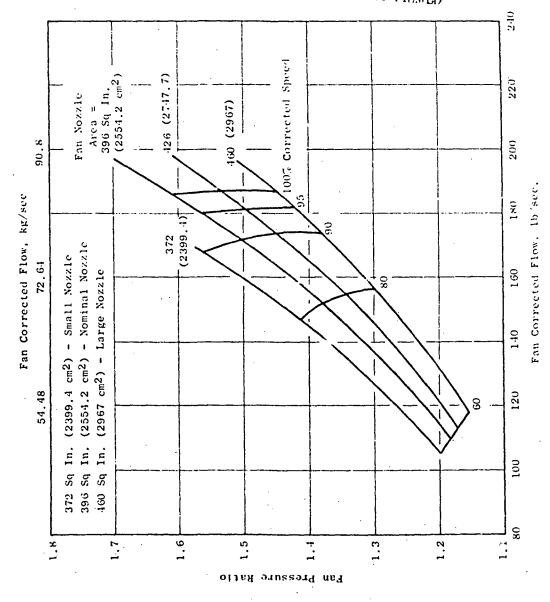


Figure 1, Performance Map, Baseline Bellmouth Inlet.

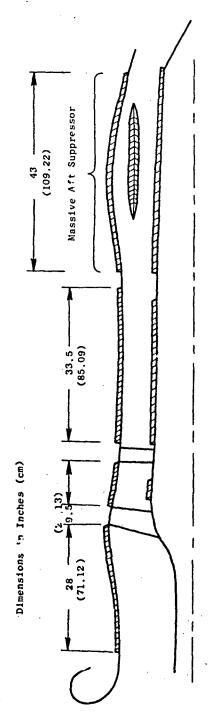


Figure 2. Fan Vehicle Cross Section.

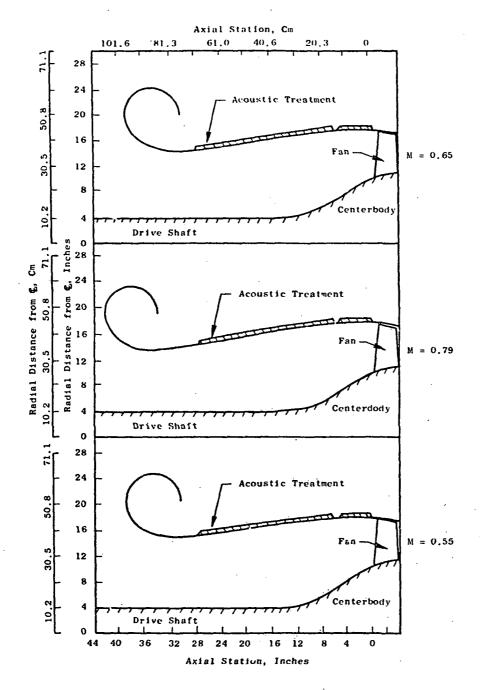


Figure 3. Inlets Without Splitters.

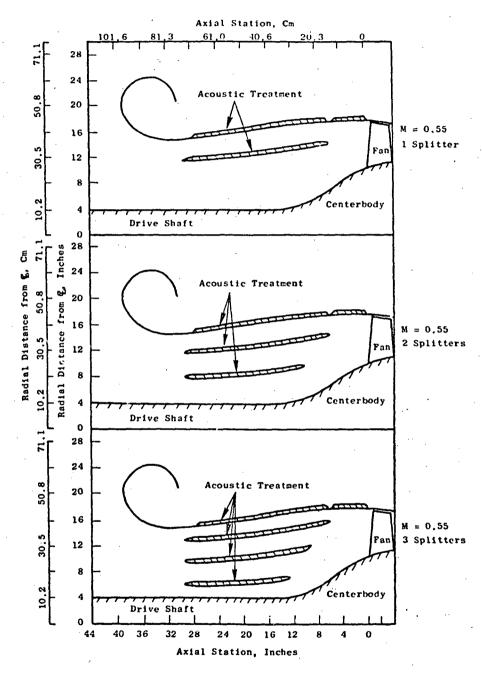


Figure 4. Inlets with Varying Splitters.

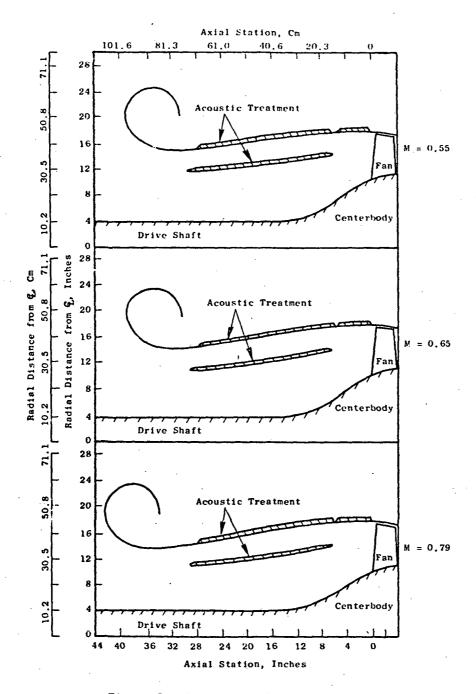
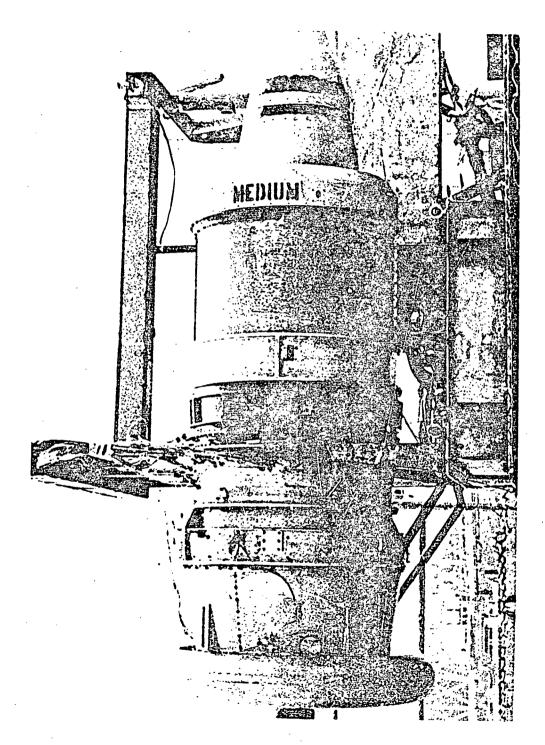


Figure 5. Inlets with One Splitter.



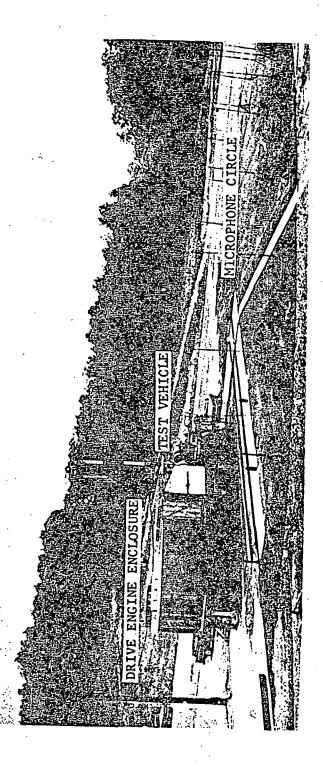


Figure 8. 200-ft (60.96 m) Sideline PNL Vs. Angle from Inlet, Takeoff.

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200-ft (60.96 m) Sideline ASPL Vs. Frequency, 70°, Takeoff, Referenced to Inlet with no Splitter. Figure 9.

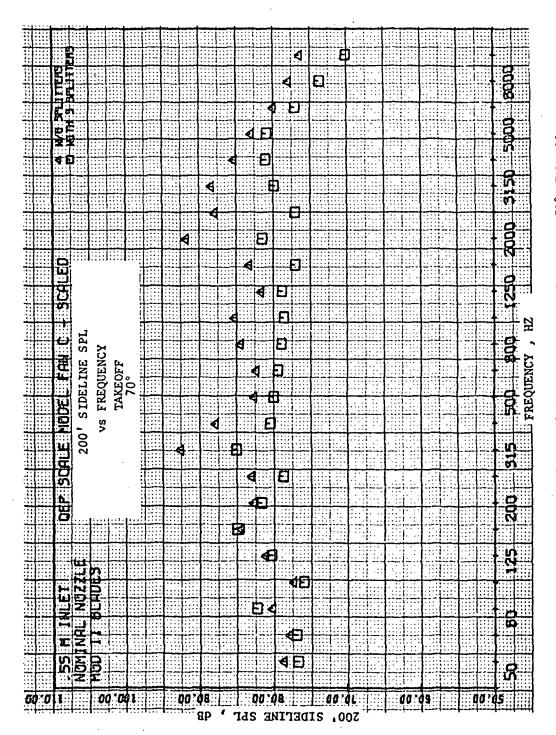


Figure 10. 200-ft (60.96 m) Sideline SPL Vs. Frequency, 70°, Takeoff.

Figure 11. 200-ft (60.96 m) Sideline SPL Vs. Frequency, 120°, Takeoff.

200-ft (60,96 m) Sideline SPN Vs. Angle from Inlet, 84% Fan Speed, Figure 12.

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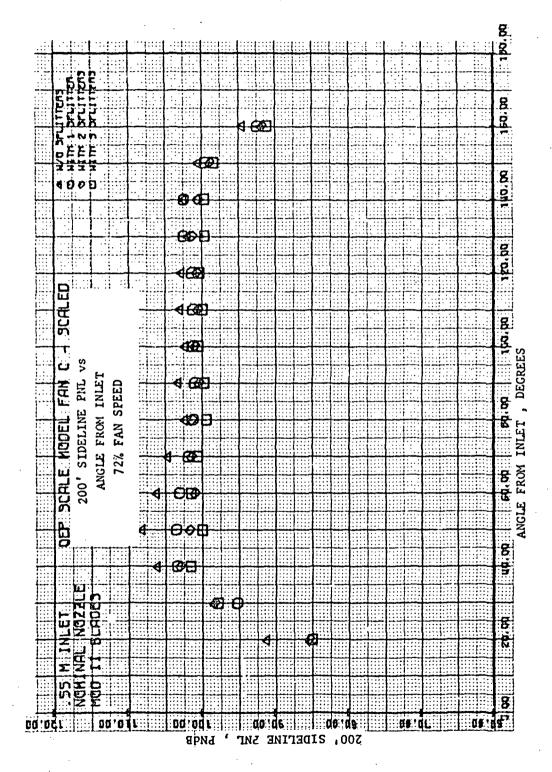
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200-ft (60.96 m) Sideline ASPL Vs. Frequency, 70°, 84% Fan Speed, Referenced to Inlet with no Splitter. Figure 13.

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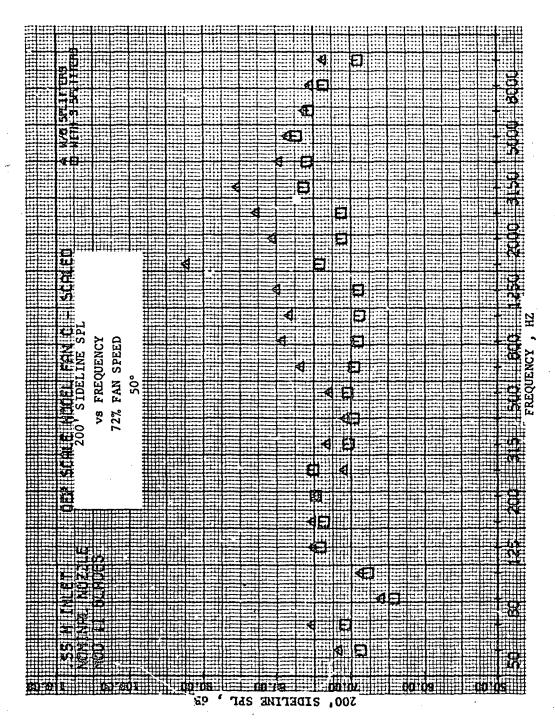
Figure 14, 200-ft (60.96 m) Sideline SPL Vs. Frequency, 70°, 84% Fan Speed.



200-ft (60,96 m) Sideline PNL Vs. Angle from Inlet, 72% Fin Speed. Figure 15.

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200-ft (60.96 m) Sideline ΔSPL Vs. Frequency, 50°, 72% Fan Speed, Referenced to Inlet with no Splitter. Figure 16.



200-ft (60.96 m) Sideline SPL Vs. Frequency, 50°, 72% Fan Speed. Figure 17.

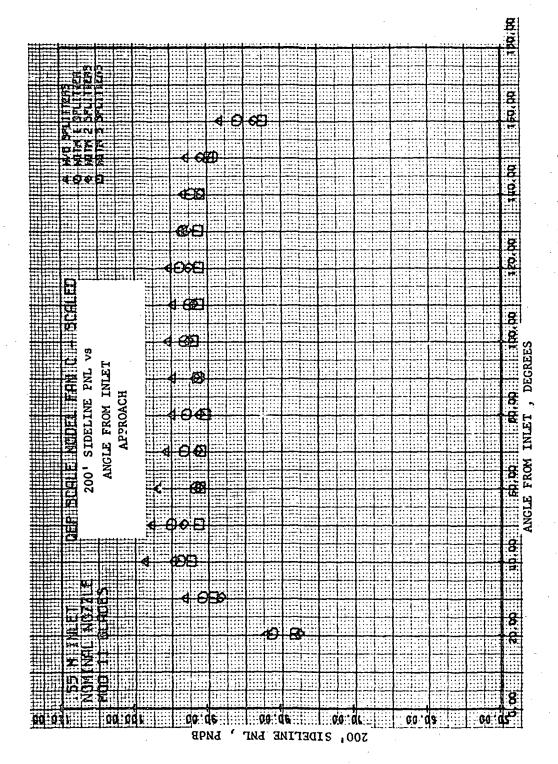
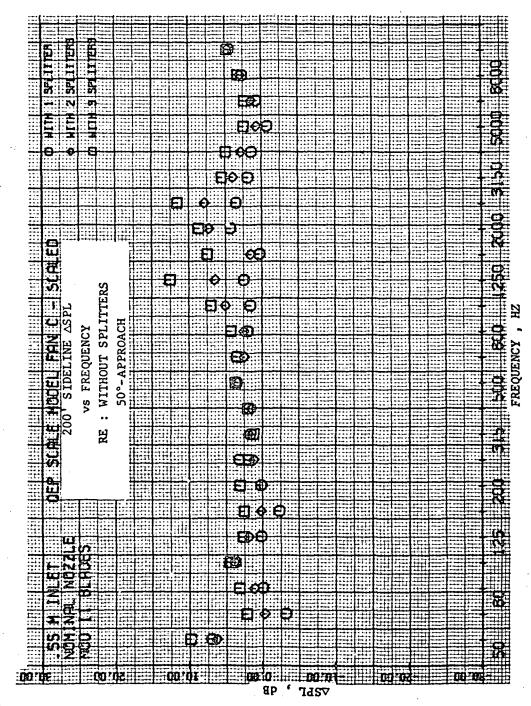
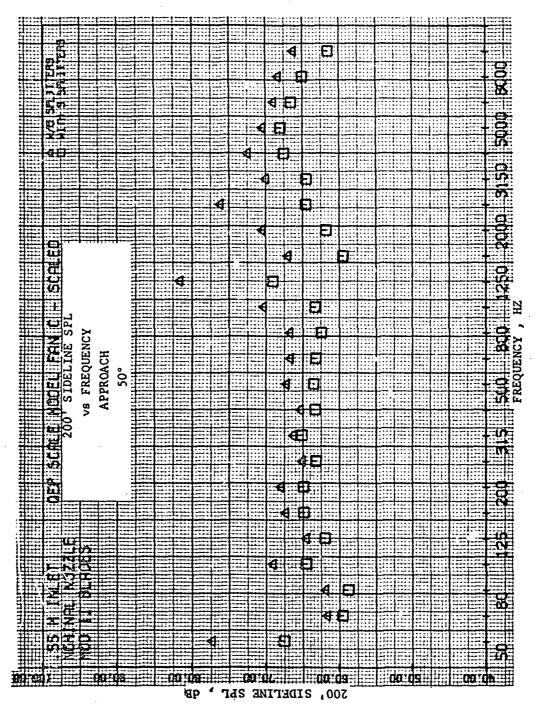


Figure 18. 200-ft (60,96 m) Sideline PNL Vs. Angle from Inlet Approach.



200-ft (60.96 m) Sideline ΔSPL Vs. Frequency, 50°, Approach, Referenced to Inlet with no Splitter. Figure 19.



gure 20. 200-ft (60.96 m) Sidleine SPL Vs. Frequency, 50°, Approach.

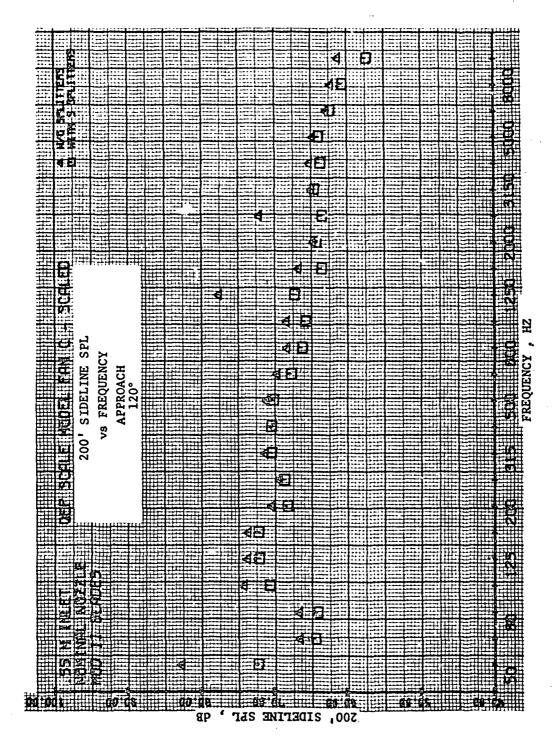


Figure 21. 200-ft (60.96 m) Sideline SPL Vs. Frequency, 120°, Approach.

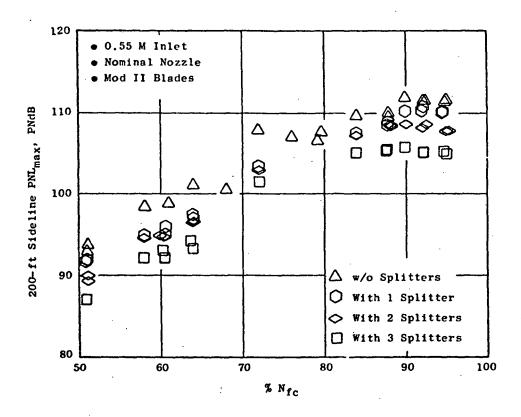
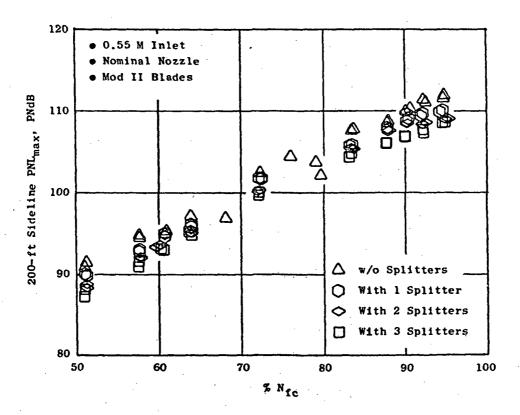


Figure 22. Forward Maximum PNL.



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Figure 23. Aft Maximum PNL.

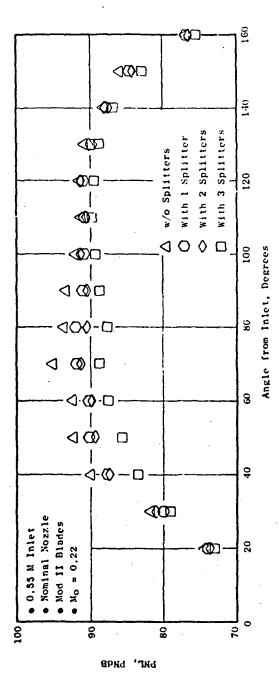
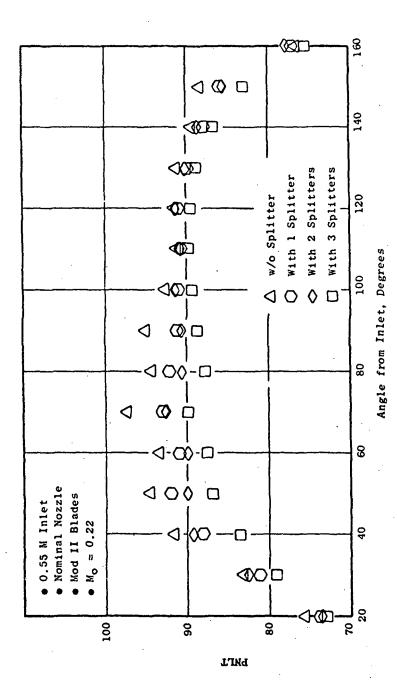


Figure 24. 1000-ft (304.8 m) Level Flyover PML, Fan plus Jet Noise (Takcoff),



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Figure 25. 1000-ft (304.8 m) Level Flyover PNLT, Fan plus Jet Noise (Takeoff).

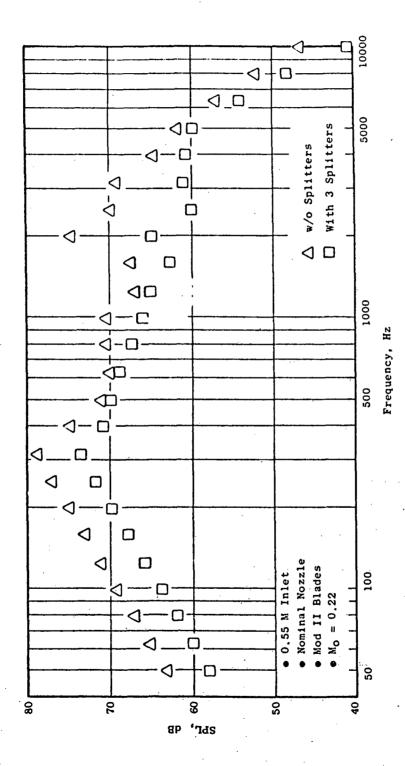
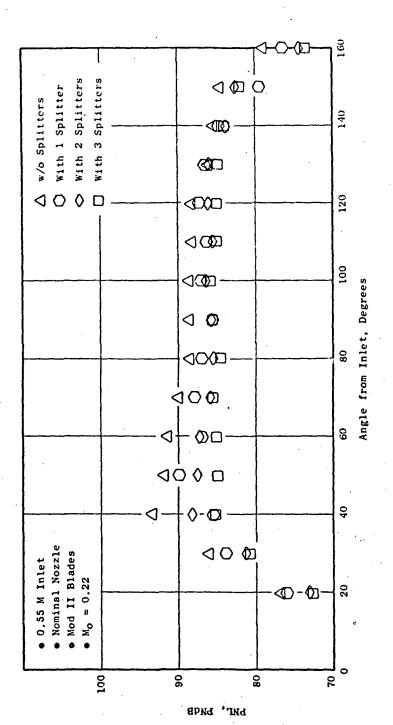


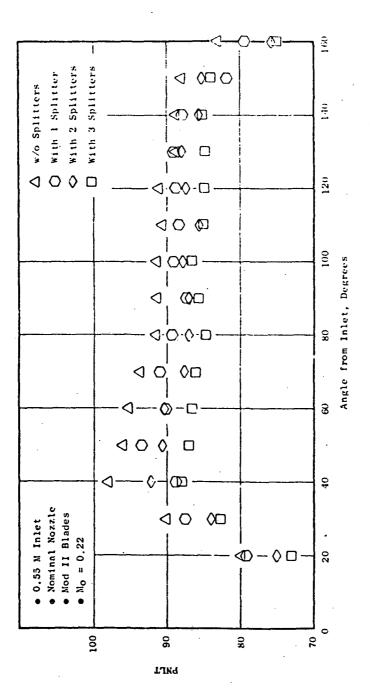
Figure 26. 1000-ft (304.8 m) Level Flyover SPL, Fan plus Jet Noise (Takcoff, 70°).



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Figure 27. 370 ft (112.8 m) Level Flyover, Fan plus Jet Noise (Approach).



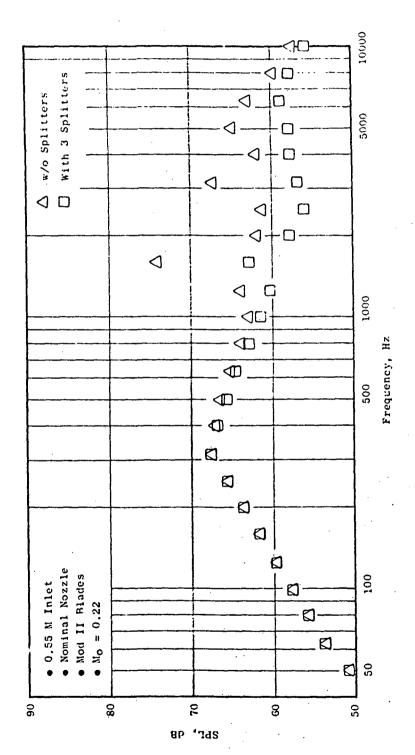


Figure 29. 370-ft (112.8 m) Level Flyover SPL, Fan plus Jet Noise (Approach, 60°).

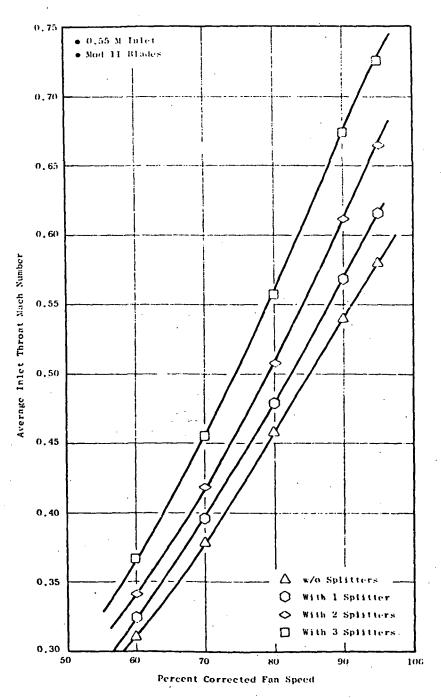


Figure 30. Average Inlet Throat Mach Number Vs. Corrected Fan Speed.

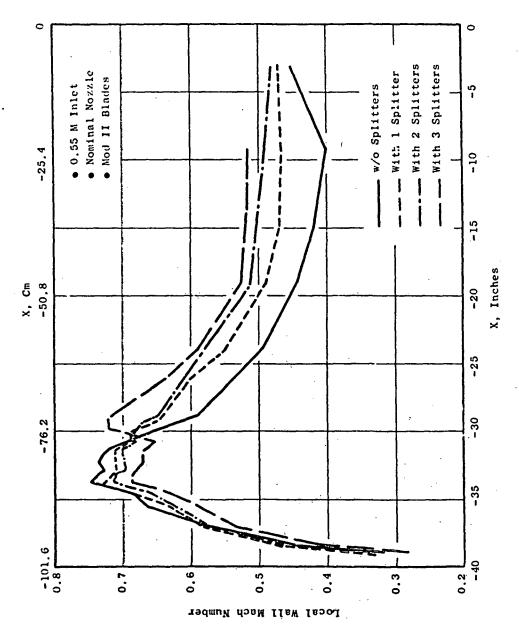


Figure 31, Outer Wall Mach Distribution, Takcoff.

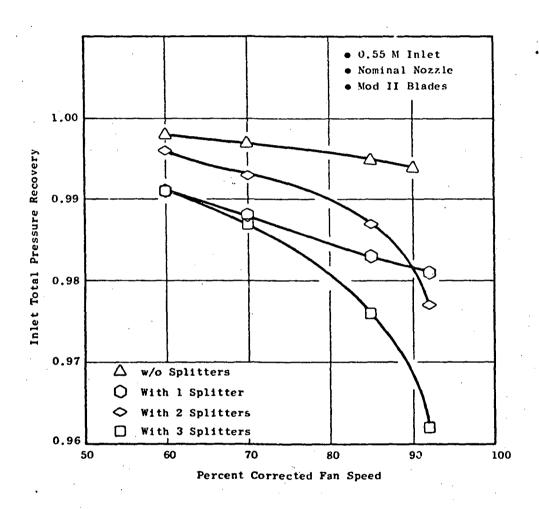


Figure 32. Inlet Total Pressure Recovery Vs. Corrected Fan Speed.

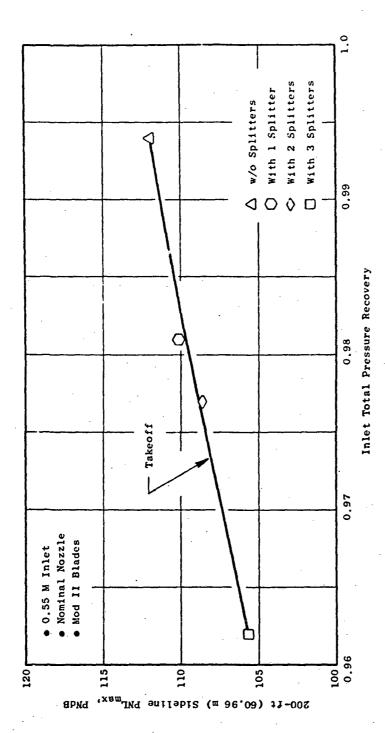


Figure 33, 200-ft (60,96 m) Sideline Front Maximum PNL.

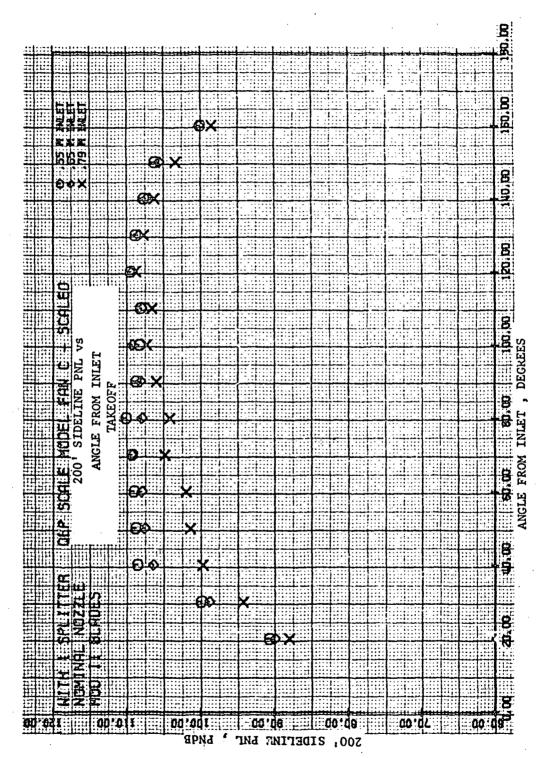
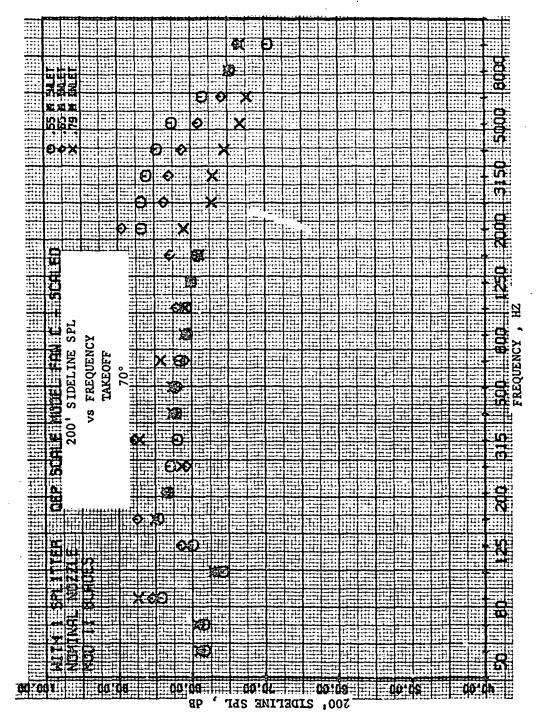


Figure 34. 200-ft (60.96 m) Sideline PNL Vs. Angle from Inlet, Takeoff.



Pigure 35, 200-ft (60,96 m) Sideline SPL Vs. Frequency, Takeoff, 70°.

200-ft (60.96 m) Sideline PNL Vs. Angle from Inlet, 84% Fan Speed.

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Figure 37, 200-ft (60,96 m) Sideline PNL Vs. Angle from Inlet, 72% Fan Speed.

Figure 38, 200-ft (60.96 m) Sideline PNL Vs. Angle from Inlet, Approach.

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Figure 39, 200-ft (60.96 m) Sideline SPL Vs. Frequency, Approach, 70°.

Figure 40. 1000-ft (304.8 m) Level Flyover PML, Fan plus Jet Noise (Takeoff).

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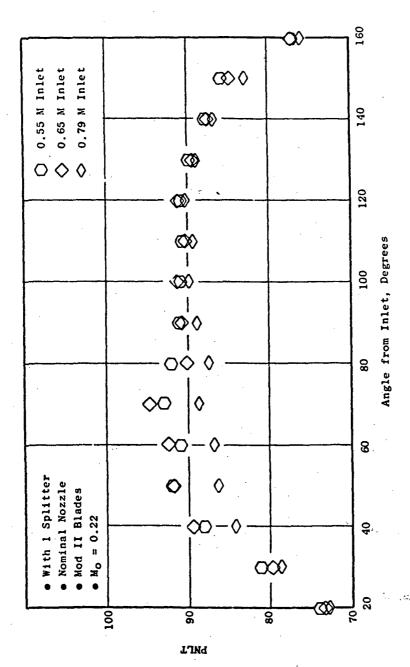
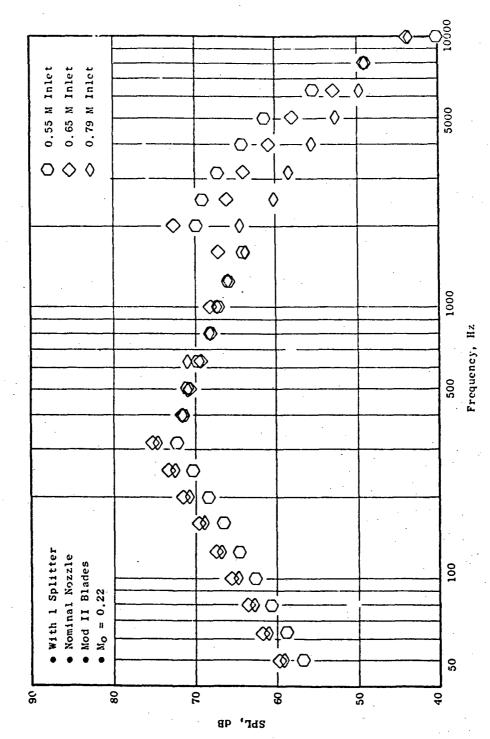


Figure 41. 1000-ft (304.8 m) Level Flyover PNLT, Fan plus Jet Noise (Takeoff).



1000-ft (304.8 m) Level Flyover SPL, Fan plus Jet Noise (Takeoff, 70:) Figure 42.

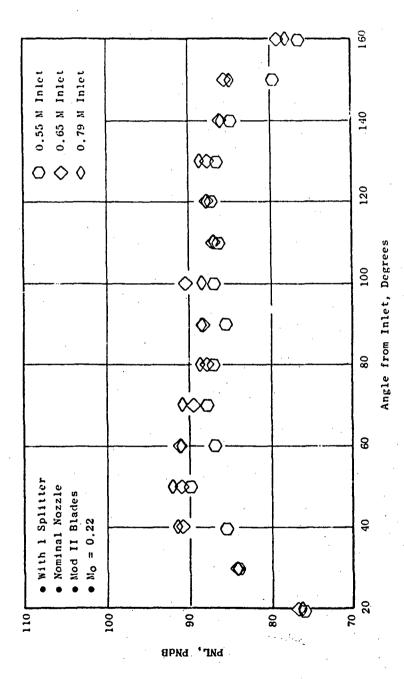
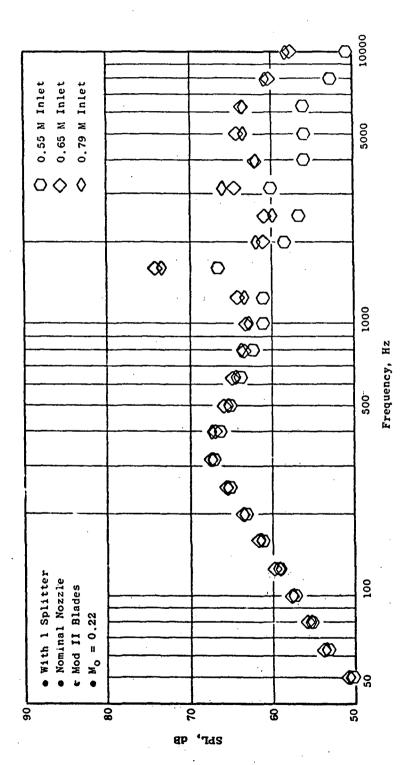


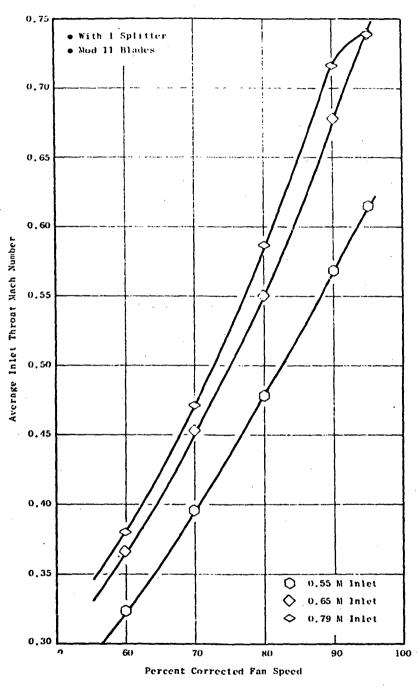
Figure 43, 370-ft (112,8 m) Level Flyover PNL, Fan plus Jet Noise (Approach).

Figure 44. 370-ft (112.8 m) Level Flyover PNLT, Fan plus Jet Noise (Approach).

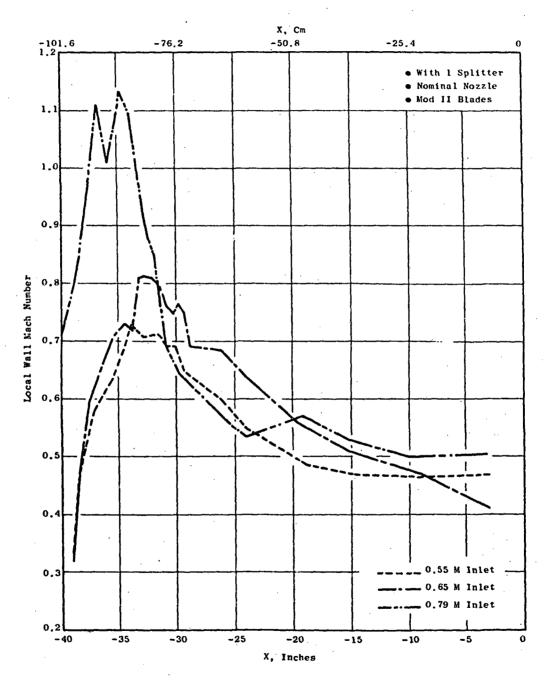


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370-ft (112.8 m) Level Flyover SPL, Fan plus Jet Noise (Approach, 60°). Figure 45.



Pigure 46. Average Throat Mach Number Vs. Corrected Fan Speed for Inlets with One Splitter.



Pigure 47. Outer Wall Mach Distribution, Takeoff.

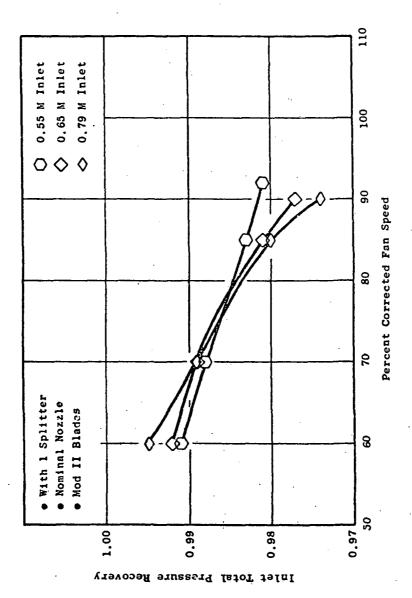


Figure 48. Inlet Total Pressure Recovery Vs. Corrected Fan Speed.

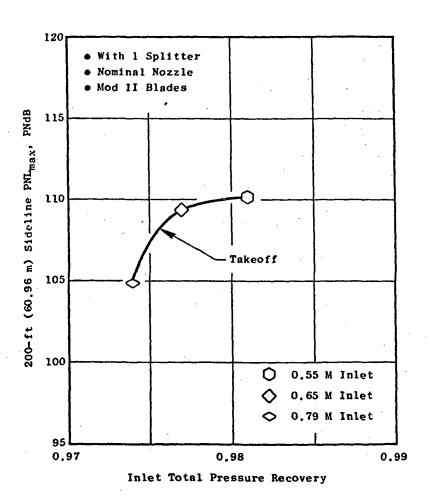
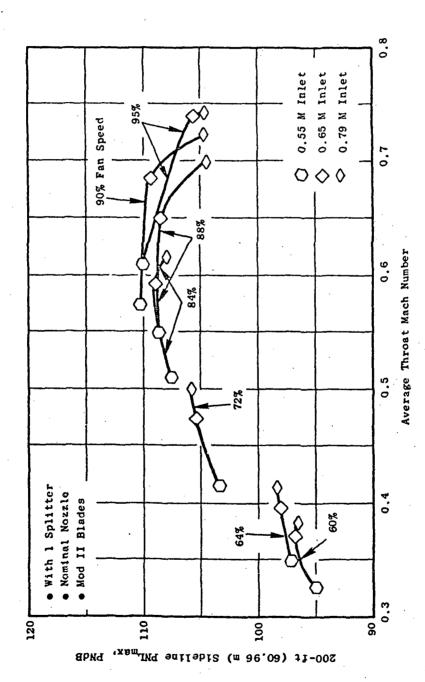


Figure 49. 200-ft (60.96 m) Sideline Front Maximum PNL, 90% Fan Speed.



200-ft (60.96 m) Sideline Front Maximum PNL, Various Fan Speeds. Figure 50.

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Figure 51, 200-ft (60.96 m) Sideline PNL Vs. Angle from Inlet, 88% Fan Speed.

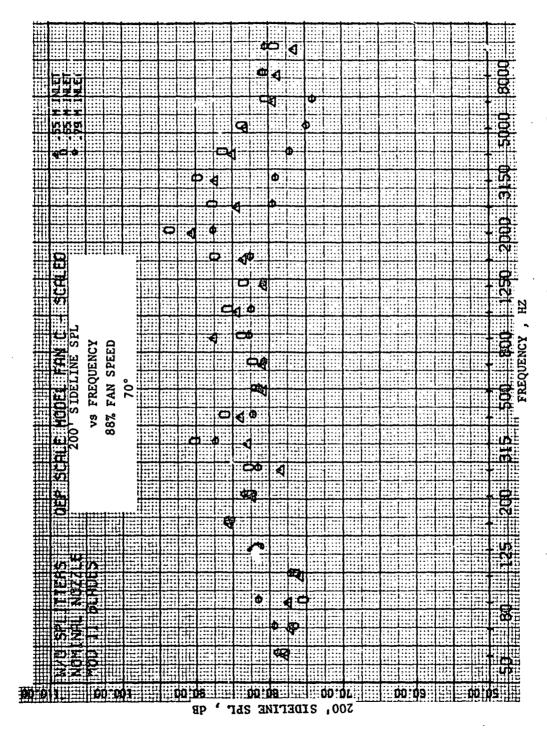
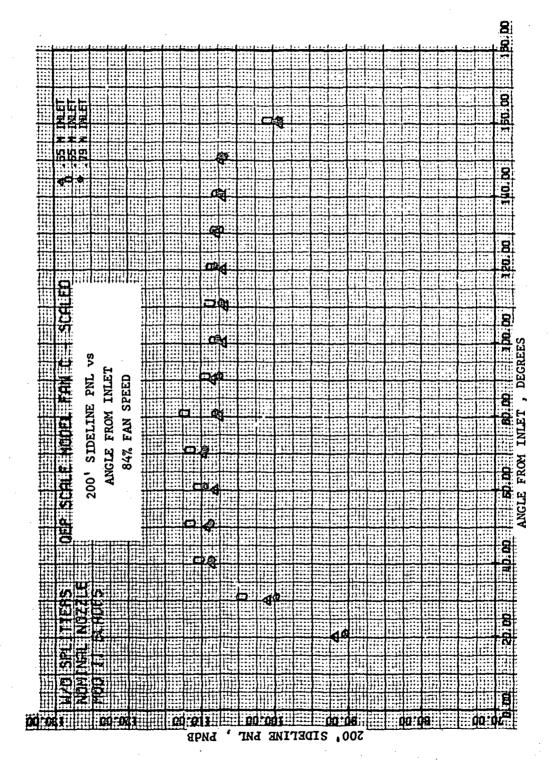


Figure 52. 200-ft (60.96 m) Sideline SPL Vs. Frequency, 88% Fan Speed, 70°



200-ft (60.96 m) Sideline PNL Vs. Angle from Inlet, 84% Fan Speed. Figure 53.

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Figure 54, 200-ft (60.16 m) Sideline PML Vs. Angle from Inlet, 72% Fan Speed.

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Pigure 55, 200-ft (60.96 m) Sideline PNL Vs. Angle from Inlet, Approach.

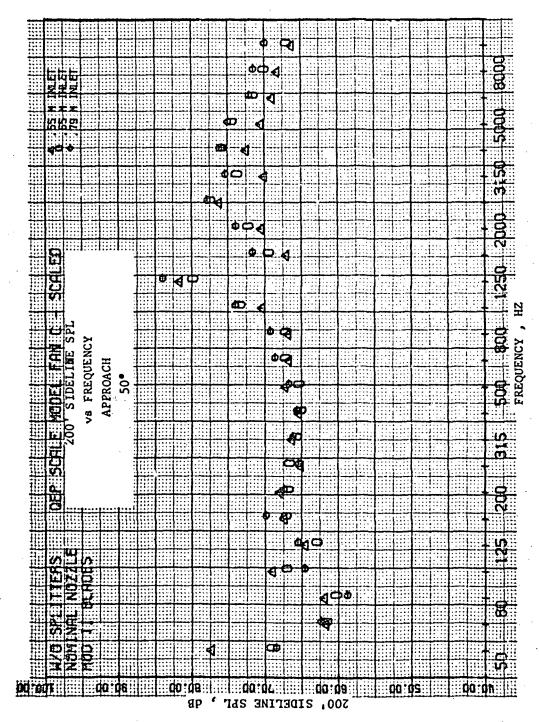


Figure 56. 200-ft (60,96 m) Sideline SPL Vs. Frequency, Approach, 50°.

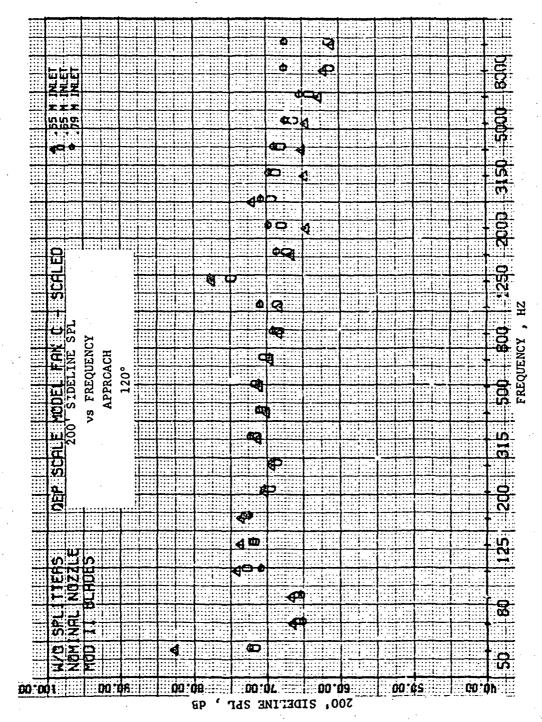
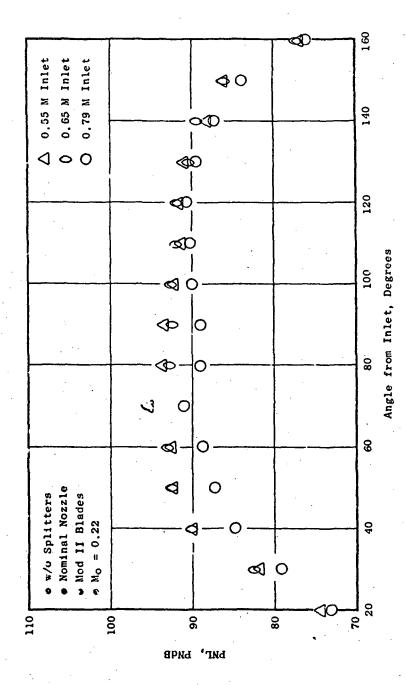
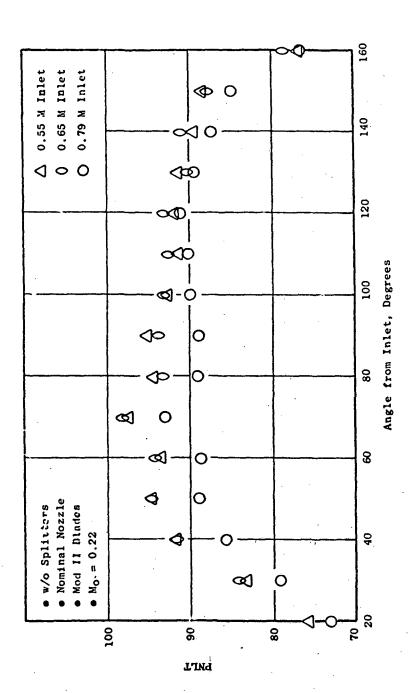


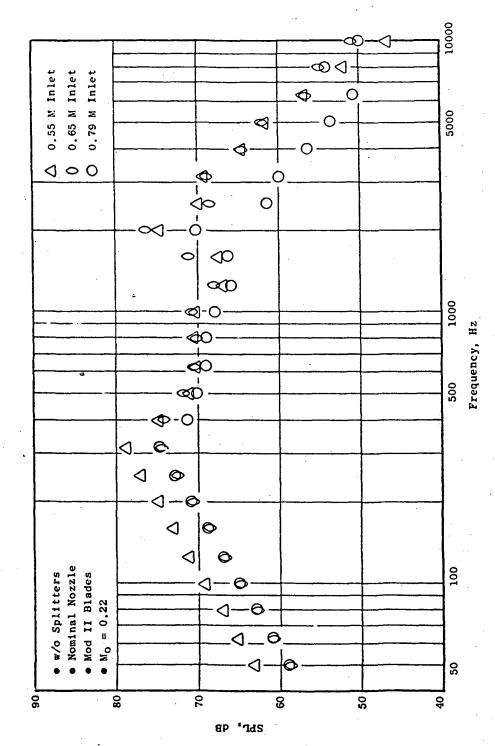
Figure 57, 200-ft (60.96 m, Sideline SPL Vs. Frequency, Approach, 120°.



1000-ft (304.8 m) Level Flyover PNL, Fan plus Jet Noise, 88% Fan Speed. Figure 58.



1000-ft (304.8 m) Level Flyover PNLT, Fan plus Jet Noise, 88% Fan Speed. Figure 59.



1000-ft (304.8 m) Level Flyover SPL, Fan plus Jet Noise, 88% Fan Speed, 70°. Figure 60.

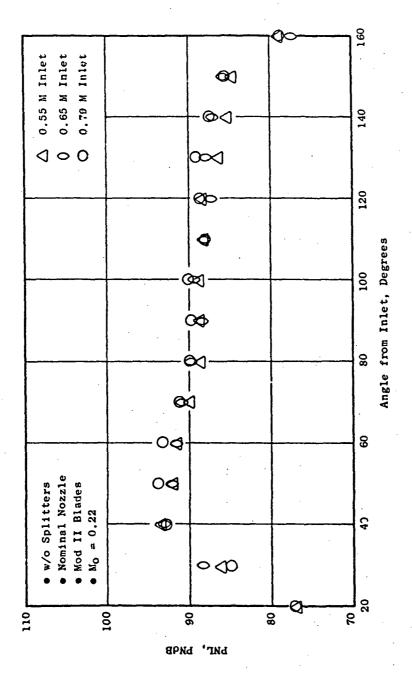
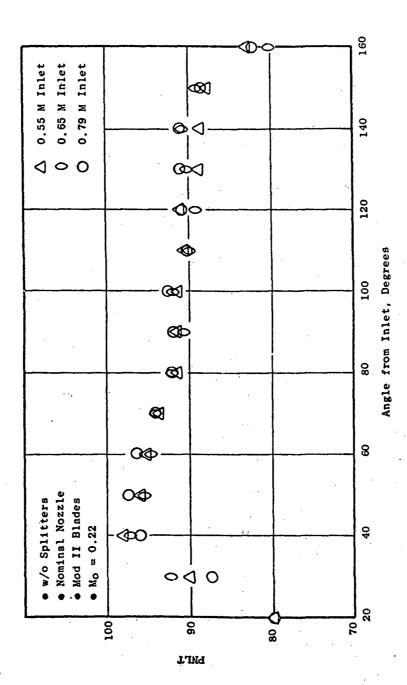


Figure 61. 370-ft (112.8 m) Level Flyover PNL, Fan plus Jet Noise, Approach.



370-ft (112.8 m) Level Flyover PNLT, Fan plus Jet Noise, Approach.

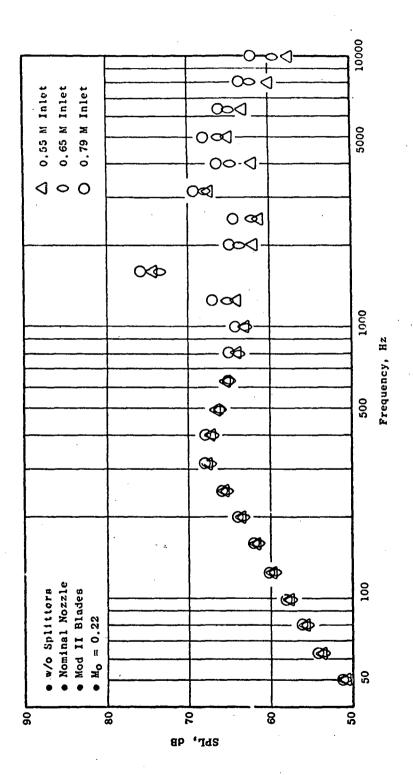


Figure 63, 370-ft (112.8 m) Level Flyover SPL, Fan plus Jet Noise, Approach, 60°.

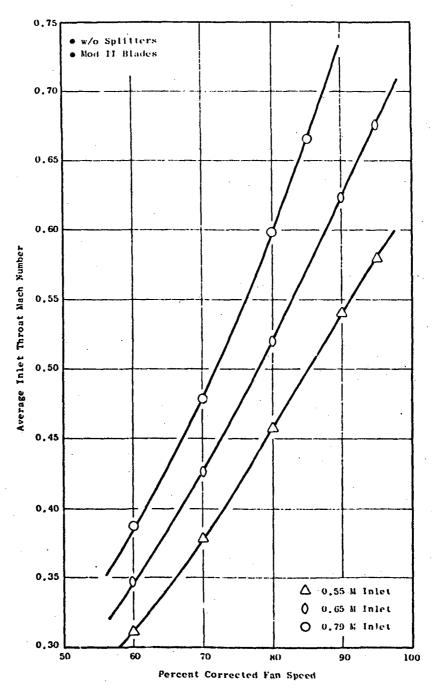
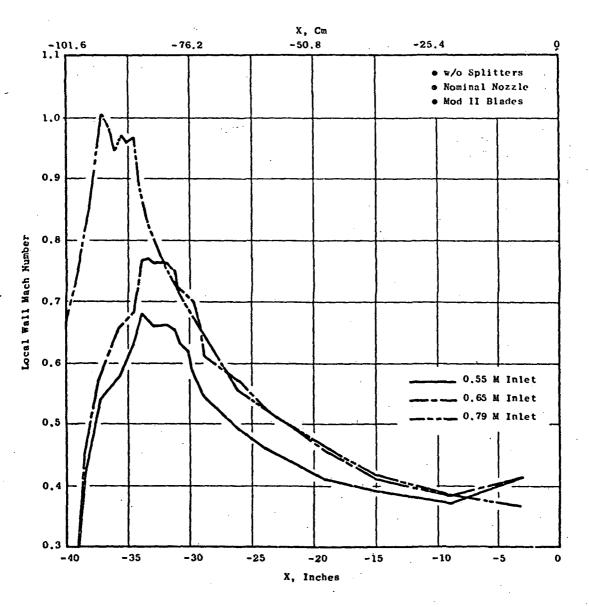


Figure 64. Average Inlet Throat Mach Number Vs. Corrected Fan Speed for Inlets Without Splitter..



Pigure 65. Outer Wall Mach Distribution, 85% Fan Speed.

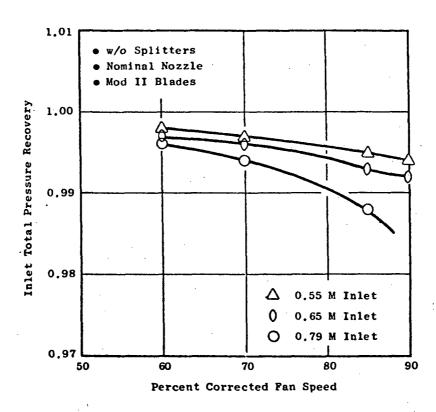


Figure 66. Inlet Total Pressure Recovery V. Corrected Fan Speed.

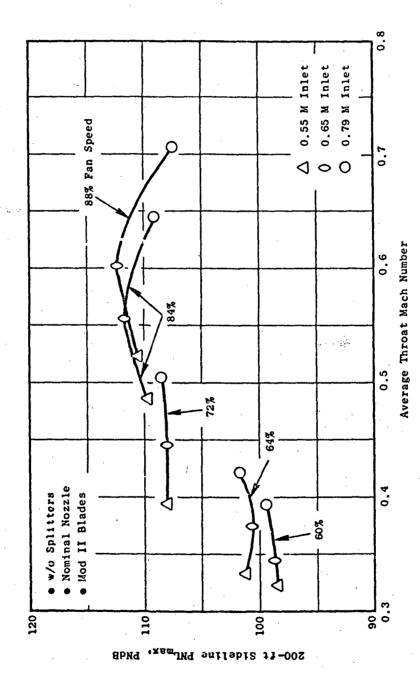


Figure 67, 200-ft (60.96 m) Sideline Front Maximum PNL.

APPENDIX B - ONE-THIRD OCTAVE DATA

This Appendix contains 1/3-octave data for high and low speed for each inlet. The data have been corrected to standard day, 59° F, 70% relative humidity. Data scaled to full scale are presented on the 200-foot (60.96 m) sideline and for reference scale model data is presented on a 100-foot (30.48 m) arc.

QEP FAN C SCALE MODEL
.55 M INIET
WITHOUT SPLITTERS
907 PAN SPEED
100' ARC

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QEP PAN C FULL SCALE
.55 M INLET
WITHOUT SPLITTERS
90% FAN SPEED
200' SIDELINE

QEP FAN C SCALE MODEL.

WITHOUT SPLITTERS 58% FAN SPEED 100' ARC CAND RADIANS

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QEP FAN C FULL SCALE
.55 M INLET
WITHOUT SPLITTERS
58% FAN SPEED
200' SIDELINE

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QEP PAN C SCALE MODEL
,55 M INLET
WITH 1 SPLITTER
90% FAN SPEED
100' ARC

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QEP FAN C FULL SCALE
.55 M INLET
WITH 1 SPLITTER
?\C\tau\tau SPEED
.00' SIDELINE

QEP PAN C SCALE MODEL

.55 M INIET
WITH 1 SPLITTER
58% FAN SFEED
100' ARC

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QEP FAN C SCALE MODEL

WITH 2 SPLITTERS 90% FAN SPEED 100' ARC .55 M INLET

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			9	77.4	78.2	79.1	65.0	600	82.7	62,4	83,7	64.2		- 9	97.0	
3			79.4	7.	0.5	69.0	87.7	87.3	85.4	89.0	84,3	86,0		66.1	82.5	
3					15.8	79.0	90.6	63.4	85.2	66.2	88.1	69.3		4.06	67.5	
	70			10.	80.0	92.6	9.78	66.2	86.1	88.8	91.2	97.6		92.0	07,0	
27				. Y		48.0	87. A	8.5	91.1	91.2	93.0	91.9		90.9	80.9	
96	2				83.7	94.6	65.5	86.1	67.7	68.0	87.9	88.0		65.0	80,0	
	4 4		0	9 0		8.4	82.	9	84.9	84.8	87.7	99.6		63.5	80.4	
200								87.4	89.3	90.0	89.3	68.9		84,7	78.4	
N 10 1				90		2	4		87.2	87.0	89.0	87.7		32.4	70,2	
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2150	201			-	200	20		90	70.0	80.4	2	70.7		74.3	65.0	
	2				10	1	70.0	77.7	70.1	78.9	79.2	78.5		711.7	64,2	
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QEP FAN C SCALE MODEL

WITH 2 SPLITTERS S8% PAN SPEED .55 M INLET

ARC 1001

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HUM, DAY) 10000 CAERALL CALCULATED PNOS QEP FAN C SCALE MODEL
.55 M INLET
WITH 3 SPLITTERS
90% FAN SPEED
100' ARC

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QRP PAN C FULL SCALE

.53 m inlet With 3 splitters 90% fan speed 200' sideline

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CAND RADIANS

PERCENT 20 HOURE DVERALL CALCUL QEP FAN C FULL SCALE
"55 M INUET
WITH 3 SPLITTERS
58% FAN SPEED
200' SIDELINE

DAYJ 10000 OVFRALL CALGULATED PWTG QEP PAN C SCALE MODEL
.65 M INLET
WITHOUT SPLITTERS
90% FAN SPEED

100' ARC

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QEP PAN C SCALE MODEL

.65 M INLET WITHOUT SPLITTERS 58% FAN SPEED 100' ARC

QEP FAN C FULL SCALE
.65 M INLET
WITHOUT SPLITTERS
58% PAN SPEED
200' SIDELINE

QEP FAN C SCALE HODEL
.65 M INLET
WITH 1 SPLITTER
90% FAN SPEED
100' ARC

CAND RADIANS) ź GEP PAN C PULL SCALE
.65 M INIZT
WITH 1 SPLITTER
90% PAN SPEED
200' SIDELINE

OEP PAN C SCALE HOUSE.

WITH 1 SPLITTER 58% PAN SPEED .65 M INLET

100' ARC

CAND RABIANS

QEP PAN C FULL SCALE
.65 M INLET
WITH 1 SPLITTER
58% PAN SPEED
200' SIDELINE

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QEP FAN C SCALE MODEL.

.79 M INLET WITHOUT SPLETTERS 86% FAN SPEED 100' ARC

QEP PAN C FULL SCALE
.79 M INLET
WITHOUT SPLITTERS
887 FAN SPEED
200' SIDELINE

QEP PAN C SCALR MODEL .79 M INLET WITHOUT SPLITTERS 58% PAN SPEED 100' ARC

QRP PAN C FULL SCALE .79 M INLET WITHOUT SPLITTERS 56% FAN SPEED 200' SIDELINE

JP M INLET WITH 1 SPLITTER 907 FAN SPEED

100' ARC

CAND RADIANS) 90 PERCENT OVERALL CALĞULATED PAGE TO SECURE A CONTRACT PROCESSOR OF THE SECURE OF THE S

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OEP FAN C SCALE HOUEL .79 M INLET

WITH 1 SPLITTER 58% PAN SPEED 100" ARC

QEP PAN C FULL SCALE
.79 M IMLET
WITH 1 SPLITTER
58% PAN SPRED
200' SIDELINE

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